

APPLICATION UNDER UNITED STATES PATENT LAWS

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Invention: MAIN GAS DUCT INTERNAL SEAL OF A HIGH-PRESSURE TURBINE

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This is a:

- ☐ Provisional Application
- ☒ Regular Utility Application
- ☐ Continuing Application
 - ☐ The contents of the parent are incorporated by reference
- ☐ PCT National Phase Application
- ☐ Design Application
- ☐ Reissue Application
- ☐ Plant Application

SPECIFICATION

MAIN GAS DUCT INTERNAL SEAL OF A HIGH-PRESSURE TURBINE

This application claims priority to German Patent Application DE10318852.5 filed
5 April 25, 2003, the entirety of which is incorporated by reference herein.

BACKGROUND OF THE INVENTION

This invention relates to a high-pressure turbine of a gas-turbine engine.

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More particularly, this invention relates to a high-pressure turbine of a gas-turbine engine with a main gas duct internal seal. Specifically, a turbine disk which carries the rotor blades is provided. Furthermore, the high-pressure turbine comprises, as usual, stator blades attached to a stator ring. A lateral wheel cavity is formed
15 between the turbine disk and the stator ring.

On high-pressure turbines, the lateral wheel cavity must always be sealed appropriately against the main gas duct to enable cooling air for the cooling of the rotor blades to be fed via the lateral wheel cavity. The known designs of sealing
20 arrangements have different overlaps between the platforms of the rotor blades and those of the stator blades. Here, the size of the gaps is such that the pressure of the main gas duct is essentially imparted to the lateral wheel cavity. With simple seal designs, the pressure disadvantageously is incapable of feeding a leading-edge cooling film for the rotor blades. To overcome these disadvantages, solutions are
25 provided in the state of the art in which the lateral wheel cavity is divided into individual chambers by means of different seals. In these solutions, the mass flow for blade cooling is usually provided from a radial inner chamber, while a radial outer chamber primarily serves to prevent hot gas from flowing in from the main gas duct.

Here, it is disadvantageous that, in particular, the outer chamber, which is located at one of the hottest zones of the high-pressure turbine outside the main gas duct, is flushed by only a very low mass flow. Furthermore, this mass flow is additionally heated on its passage through a seal. All these factors result in a high thermal load of the turbine disk (disk rim of the first stage of the high-pressure turbine). This, in turn, entails a severe limitation of the life of the turbine disk.

BRIEF SUMMARY OF THE INVENTION

In a broad aspect, the present invention provides a high-pressure turbine of a gas-turbine engine of the type specified above which avoids the disadvantages of the state of the art and provides for adequate blade cooling, while being simply designed, inexpensively producible and highly reliable in operation.

It is a particular object of the present invention to provide solution to the above problems by the combination of the features described herein. Further advantageous embodiments of the present invention will be apparent from the description below.

The present invention, accordingly, provides for the arrangement of an efficient seal between stator blade platforms and rotor blade platforms in which the rotor blade platform is used as a seal runner. The seal extends essentially in the axial direction and is located radially outward adjacent to the main gas duct.

The seal can be of the labyrinth or brush-type. In the first case, the rotor blade platforms can carry the labyrinth tips and the stator blade platforms the stationary counterpart. If the seal is of the brush-type, the brush element is, for example, attached to the stator blade platform and, in this case, can be segmented.

The design according to the present invention enables the pressure in the lateral wheel cavity, in particular in the radial outward zone of the lateral wheel cavity, to be increased in relation to the main gas duct. This enhances the supply of cooling air to

the rotor blades. In turn, the thermal situation is improved, enabling the overall temperature of the air in the rotating system to be lowered. In consequence, the temperature of the turbine disk, in particular in the disk rim area, can be lowered considerably. This results in a lower thermal load of the turbine disk and,
5 consequently, an increase of its life.

In addition, pre-swirl nozzles can be provided radially further outward. Thus, temperature reduction of the cooling air by the pre-swirl effect is further improved.

10 BRIEF DESCRIPTION OF THE DRAWINGS

The present invention is more fully described in the light of the accompanying drawings showing preferred embodiments. In the drawings,

15 Fig. 1 (Prior Art) is an embodiment according to the state of the art,

Fig. 2 is a partial longitudinal section, analogically to Fig. 1, of an embodiment in accordance with the present invention, and

20 Fig. 3 is a partial longitudinal section, analogically to Figs. 1 and 2, of a second embodiment in accordance with the present invention.

DETAILED DESCRIPTION OF THE INVENTION

In accordance with the present invention, a turbine disk 1 is provided which carries
25 rotor blades 2 of which each has a rotor blade platform 7. The turbine disk 1 rotates around a schematically shown center axis 10. Upstream of the rotor blades, stator blades 3 are arranged which are mounted to a stator ring 4. The stator blades 3, as usual, comprise stator blade platforms 6. As also illustrated in Fig. 1, a seal 11 is provided which separates an outer chamber 5b from an inner chamber 5a. Both
30 chambers 5a, 5b together form a lateral wheel cavity 5.

As to the further design details, reference is made to the designs known from the state of the art.

Fig. 2 shows a first embodiment according to the present invention, with same parts
5 being given the same reference numerals.

As becomes apparent from Fig. 2, in particular, the lateral wheel cavity 5 is separated from the main gas duct 9 by means of a radially outward seal 8. This results in a one-piece form of the lateral wheel cavity 5. Since this lateral wheel
10 cavity 5 is not divided by additional seals or the like, pre-swirl nozzle and cooling air entry into the rotor blades can be arranged on one radius and losses minimized.

In the first embodiment shown in Fig. 2, the seal according to the present invention is of the labyrinth-type and arranged between the stator blade platform 6 and the rotor
15 blade platform 7. It essentially lies on a constant radius, i.e. the seal extends on an essentially concentric circle and is not divided into several, radially offset seals, as known from the state of the art.

Fig. 3 shows a second embodiment according to the present invention which is
20 essentially designed similarly to the first embodiment shown in Fig. 2. Other than the first embodiment, the seal 8 is here of the brush type. Again, the seal lies essentially on a concentric circle, i.e. it is not radially stepped or otherwise offset, as known from the state of the art. The advantages of the design shown in Fig. 3 are similar to those of the embodiment according to Fig. 2.

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List of reference numerals

- | | |
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| 1 | Turbine disk |
| 2 | Rotor blade |
| 3 | Stator blade |
| 30 4 | Stator ring |
| 5 | Lateral wheel cavity |
| 5a | Inner chamber |

- 5b Outer chamber
- 6 Stator blade platform
- 7 Rotor blade platform
- 8 Seal
- 5 9 Main gas duct
- 10 Center axis
- 11 Seal